

METHOD FOR REPAIRING GAS TURBINE ROTOR  
BLADES

ABSTRACT OF THE DISCLOSURE

A method for replacing a portion of a gas turbine engine rotor blade, the rotor blade having a contour defined by a blade first sidewall and a blade second sidewall includes cutting through the rotor blade such that a cut line extends from a leading edge of the blade to a trailing edge of the blade, and between the first sidewall and the second sidewall, removing the portion of the rotor blade that is radially outward of the cut line, and coupling a replacement blade portion to remaining blade portion such that the newly formed rotor blade is formed with a pre-determined aerodynamic contour.